

**AMENDMENTS TO THE CLAIMS:**

This listing of claims will replace all prior versions, and listings, of claims in the application:

**Listing of Claims:**

Claim 1 (Currently amended): In a process for producing a high temperature stable fiber composite ceramic by chemical vapor infiltration (CVI) with a silicon carbide precursor in a suitable carrier gas on fiber scrims of carbon fiber preforms or silicon carbide fiber preforms, wherein methyltrichlorosilane (MTS) in hydrogen (H<sub>2</sub>) as carrier gas is used for silicon carbide deposition, the partial pressure ratio of hydrogen to methyltrichlorosilane is adjusted between 4 and 8, and the process further comprising adjusting the process pressure to  $\geq$  0.6 bar absolute and of adjusting the process temperature to  $> 1100^{\circ}\text{C}$ .

Claim 2 (Original): The process according to claim 1, wherein the process pressure is adjusted to between 0.6 bar absolute and 1.25 bar absolute and the process temperature is adjusted to between 1100°C and 1200°C.

Claim 3 (Original): The process according to claim 1, wherein the silicon carbide precursor is selected from chloro(alkyl)silanes.

Claim 4 (Canceled).

Claim 5 (Canceled).

Claim 6 (Previously presented): The process according to claim 1, wherein a heat-resistant material with a large surface is arranged between a gas feed in the reaction space and between the fiber scrims of carbon fiber preforms or silicon carbide fiber preforms to be infiltrated for preconditioning the process gas.

Claim 7 (Original): The process according to claim 6, wherein the heat-resistant material with a large surface is a carbon fiber felt.

Claim 8 (Original): The process according to claim 1, wherein carbon fiber scrims are used.

Claim 9 (Previously presented): The process according to claim 1, wherein a residual porosity is adjusted to be between from 12 to 14 percent of volume.

Claim 10 (Original): The process according to claim 1, wherein the carbon fiber preforms or silicon carbide fiber preforms are generated in that fiber layers are first constructed, the fiber layers are fixed one above the other at a distance from one another by binders, possibly accompanied by simultaneous molding and stabilization of the preform approximating the desired end product.

Claim 11 (Previously presented): The process according to claim 1, The process according to claim 10, wherein the fiber layers are generated with a 0° or 90° laying angle to a main axis of the preform.

Claim 12 (Original): The process according to claim 10, wherein the binder is selected from organic and/or silicon-organic polymer resins.

Claim 13 (Original): Structural component parts, particularly for aircraft and spacecraft, produced according to the process of claim 1.

Claim 14 (Original): Structural component parts for control flaps, leading edges of blades, nose cones, control rudders or heat protection paneling for orbital gliders and hypersonic aircraft produced according to the process of claim 1.